

there must be many intersections with so high a porosity, and the Johnson-Stewart model neglects the volume of these intersections.

It is concluded that application of the Johnson-Stewart method for predicting diffusion rates may be hazardous for porous materials either containing abnormally small pores or having abnormally high porosities. For materials having pore structures of types more frequently encountered, the behavior of the Harshaw alumina reinforces previous conclusions of reasonable a priori predictability and good extrapolation properties with pressure.

It may also be noted that the effect of temperature on the predictive capability of the Johnson-Stewart method was very slight.

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Impact Tube Gas Velocity Measurement at High Temperatures

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Impact tubes (Pitot tubes) are frequently used to determine the line or velocity of a flowing fluid at a point. The inviscid model, or Bernoulli equation, can be successfully applied in gas flows at reasonable temperatures, but when extremely high temperatures are encountered this model becomes suspect. Because gas viscosities increase with temperature, and because of compressibility effects, the usual impact tube results no longer apply.

When temperatures exceed those normally found in flames (1,000-2,000 K), such conditions may be encountered. The impact tube must then be internally cooled, or destruction will result. Thus the phenomenon of heat transfer also enters the picture. A model of an impact tube must correctly predict both heat and momentum transfer when the cooled tube is immersed in an extremely hot gas (plasma).

STAGNATION STREAMLINE FLOW

For isentropic, inviscid deceleration of an incompressible fluid along a stagnation streamline, the stagnation pressure P_0 as measured with an impact (Pitot) tube is

given in terms of the free stream quantities—static pressure P_∞ , density ρ_∞ , and velocity U —by the Bernoulli equation

$$P_0 = P_\infty + \frac{\rho_\infty U^2}{2} \quad (1)$$

If the fluid is compressible, the equation for isentropic, inviscid deceleration becomes

$$P_0 = P_\infty \left(1 + \frac{\gamma - 1}{2} M_\infty^2 \right)^{\frac{\gamma}{\gamma - 1}} \quad (2)$$

where M_∞ is the free stream Mach number and γ is the ratio of specific heats C_p/C_v .

For viscous fluids flowing at Reynolds numbers above 100 (the characteristic length chosen to be the radius of the impact tube), the viscous forces acting on the fluid are very small in comparison to the inertia forces and the above equations are valid. As the Reynolds number is decreased below 100, however, the measured impact pressure departs from the true stagnation pressure based on isentropic, inviscid deceleration, and it is no longer possible to ignore the effects of viscosity. This departure of the impact pressure from the Bernoulli pressure is known as the Barker (1922) effect and has been observed experimentally by Hurd (1953), MacMillian (1954), Schowalter and Blaker (1961), and Sherman (1953). It shows

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itself in the form of an additional pressure which can be calculated for simple geometries (that is, cylinders, spheres), so that the measured impact pressure is greater than the Bernoulli pressure.

Carleton (1970) has shown that the difference between the measured impact pressure and the free stream pressure $P_0 - P_\infty$ for a plasma flowing around a hemispherical tipped cylindrical body can be expressed by a Bernoulli term, a compressibility term, and a viscous term:

$$P_0 - P_\infty = \frac{\rho_\infty U^2}{2} + \frac{\rho_\infty^2 U^4}{8\gamma P_\infty} + \frac{2\bar{\mu}U}{R\left(1 + \frac{0.5576}{\sqrt{Re}}\right)} \quad (3)$$

Physical properties in the first two terms are evaluated at free stream conditions of temperature and pressure. In order to select $\bar{\mu}$ and evaluate Re in the third term, a characteristic temperature of the boundary layer is chosen. This characteristic or reference temperature is a temperature at which constant property results will closely approximate true variable property behavior. The appropriate values of $\bar{\mu}$ and Re are those corresponding to the reference temperature. Although no prior basis for the use of this technique in a plasma boundary layer has been established, it is a method which does compensate for the temperature gradient near the probe. Chambre and Smith (1949) in a similar manner developed an expression for a high velocity low density gas stream which differs somewhat from Equation (3) in the viscous term because flow around a spherical body was assumed.

STAGNATION POINT HEAT TRANSFER

The validity of using the reference temperature technique in plasma flow calculations is established by considering heat transfer to an object immersed in a plasma stream and applying the resulting theoretical expression to previously unresolved experimental data. In order to determine heat transfer rates at the stagnation point of the object, the equations describing the flow in the boundary layer surrounding the object must be analyzed.

Thus, by assuming the fluid properties constant, evaluated at the reference conditions, the system of boundary layer equations becomes

$$r \frac{\partial u}{\partial z} + \frac{\partial}{\partial r}(rv) = 0 \quad (4)$$

$$\rho^* \left(v \frac{\partial v}{\partial r} + u \frac{\partial v}{\partial z} \right) = - \frac{\partial P}{\partial r} + \mu^* \frac{\partial^2 v}{\partial z^2} \quad (5)$$

$$\rho^* (C_P + C_{Pr})^* \left(v \frac{\partial T}{\partial r} + u \frac{\partial T}{\partial z} \right) = (K + K_r)^* \frac{\partial^2 T}{\partial z^2} \quad (6)$$

In the above equations * denotes reference conditions. Sibulkin (1952), following the method of Squire, given by Goldstein (1938), and using Homann's (1952) stagnation point solution of the momentum equation, obtained a solution to the above system of equations for laminar, incompressible low speed flow. In dimensionless form the heat transfer coefficient is given by

$$h = \frac{0.763(K + K_r)^*}{R} \left[\frac{(C_p + C_{Pr})^* \mu^*}{(K + K_r)^*} \right]^{0.4} \left[\frac{\rho^* R U}{\mu^*} \right]^{0.5} \quad (7)$$

where R is the radius of the half body, U the free stream

approach velocity, and all fluid properties are evaluated at the reference temperature. This is taken to be the temperature corresponding to the mean boundary layer enthalpy:

$$h^* \equiv h_w + 0.5 (h_\infty - h_w) \quad (8)$$

VERIFICATION OF HEAT TRANSFER MODEL

To demonstrate the applicability of the reference temperature method to plasma flow problems, Equation (7) was used to resolve the data of Chludzinski (1964). Chludzinski studied heat transfer to solids immersed in an r.f. generated plasma at atmospheric pressure. Heat transfer coefficients were determined from the dynamic response of shielded thermocouples exposed to an argon plasma for time periods of approximately 0.1 sec. The thermocouple was housed within a water-cooled probe and was advanced to the probe stagnation point and retracted by means of solenoids mounted above the cooling water jacket.

The theoretical predictions of h obtained by using Equation (7) for the three probe sizes are plotted in Figure 1 as a function of the free stream value of α , the degree of argon ionization. The range of α corresponds to the free stream temperature range of 8,500 K to 10,500 K. The temperature scale, which is not linear, is indicated.

CONFIRMATION OF PRESSURE RECOVERY MODEL

To test the model for pressure recovery at the stagnation point of a cool impact probe in a plasma stream, experimental results must be considered. In a recent study of Carleton (1970), flow patterns in a confined argon plasma jet, profiles of impact pressure, and temperature were obtained in the jet at two argon flow rates. Velocity profiles were then calculated. The pressure profiles were measured with a water-cooled impact probe which moved continuously through the jet along the jet diameter. Property data were taken from Devoto (1966, 1967), Drellishak (1963), Meador and Stratton (1965) and Sherman (1963).

A check on the values obtained for the velocities in the two plasma jets is obtained by integrating the velocity profile to determine the mass flow rate. The calculated value can then be compared with the metered argon flow rate to the plasma generator. For the two jets, the veloci-

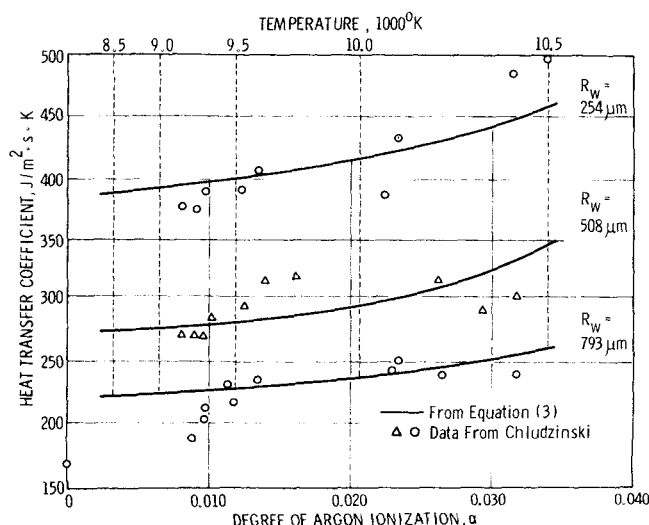


Fig. 1. Calculated and observed heat transfer coefficients as a function of the degree of argon ionization.

TABLE 1. EFFECT OF VISCOSITY AND COMPRESSIBILITY
ON PLASMA VELOCITIES

Jet position x/R	P N/m^2	$T/^\circ K$	U m/s	U_B m/s	Vis %	Com %
Low flow rate			Metered flow = .417 g/s Calculated flow = .439 g/s			
-0.948	0.057	8,320	4.08	4.40	14.1	0.1
-0.474	0.307	9,730	10.60	11.12	8.4	0.8
0.000	0.800	10,910	18.44	19.32	6.8	2.1
0.474	0.524	10,260	14.34	15.00	7.3	1.4
0.948	0.074	8,260	4.66	4.99	12.5	0.1
High flow rate			Metered flow = .602 g/s Calculated flow = .600 g/s			
-0.948	0.202	8,300	7.93	8.29	7.9	0.5
-0.474	0.533	9,780	14.10	14.69	6.6	1.4
0.000	1.080	11,000	21.56	22.58	6.0	2.8
0.474	0.752	10,470	17.43	18.21	6.4	2.0
0.948	0.097	8,200	5.38	5.71	10.9	0.2

ties at each point on the profile were multiplied by the corresponding density and area and summed. The area used for each velocity zone was that of a half annular shell of the appropriate mean radius. Since the velocities were indicated only for the luminous region which is slightly smaller than the region of momentum exchange, the temperature profiles were extrapolated linearly to calculate velocities at the fringes of the impact pressure profiles. Two additional values were obtained in this manner at each edge. The results are presented in Table 1.

The effects of compressibility and viscosity on the velocity measurements in each plasma jet, for five locations along the velocity profile, is indicated in Table 1 for two flow rates. The impact pressure, the temperature, the actual velocity, the Bernoulli velocity, and contributions of the viscosity and compressibility terms of Equation (3) to the measured impact pressure are listed.

CONCLUSIONS

A model has been developed to relate the pressure sensed by a water-cooled impact probe in a plasma flow field to the free stream plasma velocity. The model considers the effects of both viscosity and heat transfer. The model was applied to determine velocity profiles in a plasma jet. Integration of the velocity profiles yielded mass flow rates which agreed closely with the metered flow rates to the plasma generator. Heat transfer to the probe is also predicted well.

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NOTATION

C_p	= specific heat capacity at constant pressure, J/kg·K
C_{Pr}	= reactive heat capacity at constant pressure, J/kg·K
C_v	= specific heat capacity at constant volume, J/kg·K
Com	= contribution of the compressibility term in Equation (3) to the measured impact pressure, %
h	= specific enthalpy, J/kg
h	= heat transfer coefficient, J/m ² ·s·K
K	= thermal conductivity, J/m·s·K
K_r	= reactive thermal conductivity at constant pressure, J/m·s·K
M	= Mach number

P	= pressure, N/m ²
r	= coordinate normal to the stagnation streamline, m
R	= radius of curvature of a spherical half body, m
Re	= Reynolds number
T	= temperature, K
u	= component of velocity parallel to the stagnation streamline, m/s
U	= axial component of the free stream velocity, m/s
v	= component of velocity normal to the stagnation streamline, m/s
Vis	= contribution of the viscosity term in Equation (3)
z	= coordinate parallel to the stagnation streamline, m
α	= fraction of argon which is ionized
μ	= viscosity, kg/m·s
ρ	= mass density, kg/m ³
γ	= heat capacity ratio

Subscripts

0	= stagnation conditions
∞	= free stream conditions
w	= wall conditions

Superscripts

—	= mean value
*	= reference conditions

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